



318
12/6/98

IN THE UNITED STATES PATENT AND TRADEMARK OFFICE

Applicants: Mannava, et al.)
Serial No:) Group Art Unit:
Filed: HEREWITH)
For: LASER SHOCK PEENED GAS TURBINE)
ENGINE COMPRESSOR AIRFOIL EDGES)

Hon. Assistant Commissioner for Patents
Washington, D.C. 20231

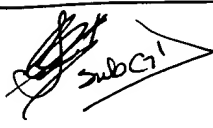
SIR:

PRELIMINARY AMENDMENT

Please enter the following Preliminary Amendment for the above-identified Continuation US patent application (the parent case of which is Serial No: 08/719,341 filed September 25, 1996 and presently under Final Rejection) filed simultaneously herewith:

IN THE CLAIMS:

Please amend the following claims as indicated.

 1. (FOUR TIMES AMENDED) A gas turbine engine component comprising:

a metallic compressor airfoil having a leading edge and a trailing edge and a pressure side and a suction side,

at least a first laser shock peened surface on a first side of said airfoil,

said laser shock peened surface extending radially along at least a portion of said leading edge and extending chordwise from said leading edge, and

a first region having deep compressive residual stresses , imparted by laser shock peening (LSP) extending into said airfoil from said laser shock peened surface wherein said deep